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Filing Date	09/18/2003
First Named Inventor	Anthony W. Russell
Art Unit	2859
Examiner Name	Bennett, George B.
Attorney Docket Number	SMAR002

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ENCLOSURES (*Check all that apply*)

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Firm Name	Mark A. Oathout
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20SEP02 E749514-1 002884
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Cardiff Road
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1. Your reference	P31613-/SGR/GWO/PPP	
2. Patent application number *(The Patent Office will fill in this part)*	0221753.7	19 SEP 2002
3. Full name, address and postcode of the or of each applicant *(underline all surnames)*	Smart Stabilizer Systems Limited Unit 600, Ashchurch Business Centre Alexandra Way Ashchurch Tewkesbury Gloucestershire GL20 8GA	
Patents ADP number *(if you know it)*	08468381001	
If the applicant is a corporate body, give the country/state of its incorporation	United Kingdom	
4. Title of the invention	"Borehole Surveying"	
5. Name of your agent *(if you have one)*	Murgitroyd & Company	
"Address for service" in the United Kingdom to which all correspondence should be sent *(including the postcode)*	Scotland House 165-169 Scotland Street Glasgow G5 8PL	
Patents ADP number *(if you know it)*	1198013	

6. If you are declaring priority from one or more earlier patent applications, give the country and the date of filing of the or of each of these earlier applications and *(if you know it)* the or each application number	Country	Priority application number *(if you know it)*	Date of filing *(day / month / year)*

7. If this application is divided or otherwise derived from an earlier UK application, give the number and the filing date of the earlier application	Number of earlier application	Date of filing *(day / month / year)*

8. Is a statement of inventorship and of right to grant of a patent required in support of this request? *(Answer 'Yes' if:* *a) any applicant named in part 3 is not an inventor, or* *b) there is an inventor who is not named as an applicant, or* *c) any named applicant is a corporate body.* *See note (d))*	Yes

9. Enter the number of sheets for any of the following items you are filing with this form. Do not count copies of the same document

Continuation sheets of this form	
Description	16
Claim *(s)*	- DMC
Abstract	-
Drawing *(s)*	~~45~~ 4

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Priority documents	-
Translations of priority documents	-
Statement of inventorship and right to grant of a patent *(Patents Form 7/77)*	-
Request for preliminary examination and search *(Patents Form 9/77)*	-
Request for substantive examination *(Patents Form 10/77)*	-
Any other documents *(please specify)*	-

11. I/We request the grant of a patent on the basis of this application.

Signature Murgitroyd & Company

Date 18/09/2002

12. Name and daytime telephone number of person to contact in the United Kingdom — Paolo Pacitti — 0141 307 8400

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Notes

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b) *Write your answers in capital letters using black ink or you may type them.*

c) *If there is not enough space for all the relevant details on any part of this form, please continue on a separate sheet of paper and write "see continuation sheet" in the relevant part(s). Any continuation sheet should be attached to this form.*

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"Borehole Surveying"

This invention relates to a method and apparatus for use in surveying of boreholes.

It is known in directional drilling, for example, to detect the orientation of a drillstring adjacent to the bit by means of a sensor package for determining the local gravitational **[GX,GY,GZ]** and magnetic **[BX,BY,BZ]** field components along mutually orthogonal axes, and to derive from these the local azimuth **(AZ)** and inclination **(INC)** of the drillstring. Conventionally, the measurements are made by providing within the instrument package three mutually perpendicular accelerometers and three mutually perpendicular magnetic fluxgates.

The present invention is concerned with an arrangement which requires only two measurement devices, namely a single accelerometer and a single magnetic fluxgate or a single accelerometer and a single rate gyro, the latter being preferred for situations in which magnetic interference is likely to be encountered.

Accordingly, the present invention provides a method of surveying boreholes, comprising:

providing an instrument package in the leading end of a drillstring, the instrument package comprising first and second single-axis sensors mounted for rotation with the drillstring about the rotational axis of the drillstring, the first sensor being an accelerometer and the second sensor being a magnetic fluxgate or a rate gyro;

rotating the drillstring;

deriving from the first sensor the inclination angle of the drillstring at the instrument package; and

deriving from the second sensor the azimuth angle of the drillstring at the instrument package.

Each of the sensors will typically be positioned in one of two configurations. In the first configuration, the sensor is radially spaced from the borehole axis and has its sensing axis in a plane containing the borehole axis and an axis perpendicular thereto. In the second configuration, the sensor is radially spaced from the borehole axis and has its sensing axis in a plane parallel with the borehole axis.

Preferably, the drilling control rotation angle is also obtained from the sensor outputs.

Preferably, the sensor outputs are integrated over the four quadrants of rotation and the desired output angle is derived from the integrated output. The instrument package suitably includes rotation angle reference means for use in the integration.

Additional information may be derived, such as the local gravitational and magnetic field vectors.

From another aspect, the invention provides apparatus for use in surveying boreholes, the apparatus comprising an instrument package adapted to be included in the leading end of a drillstring, the instrument package comprising first and second single-axis sensors mounted for rotation with the drillstring about the rotational axis of the drillstring, the first sensor being an accelerometer and the second sensor being a magnetic fluxgate or a rate gyro; and computing means for deriving from the first sensor while the drillstring is rotating the inclination angle of the drillstring at the instrument package, and for deriving from the second sensor while the drillstring is rotating the azimuth angle of the drillstring at the instrument package.

The computing means preferably operates to integrate the sensor outputs over the four quadrants of rotation and to derive the desired output angle from the integrated output.

The apparatus may further include rotation angle reference means for use in the integration.

Examples of the present invention will now be described, by way of illustration only, with reference to the drawings, in which:

Fig. 1 illustrates, in general terms, the operation of a single axis sensor in a drillstring for sensing any given vector V;

Fig. 2 is a block diagram of one circuit which may be used to identify rotation quadrant;

Fig. 3 illustrates the operation where the sensor is an accelerometer;

Fig. 4 illustrates the operation where the sensor is a fluxgate;

Fig. 5 illustrates the derivation of azimuth angle; and

Fig. 6 illustrates the operation where the sensor is a rate gyro.

<u>Single-axis sensor</u>

The operation of a single-axis sensor in a drill string will first be described in general terms. The application of this to specific sensors is discussed below.

Referring to Fig. 1, a single-axis sensor 10 is mounted on a drill string (not shown). The sensor 10 senses a fixed vector **{V}** and is mounted in one of two configurations.

In the first configuration, the sensor 10 lies in a plane containing the rotation axis **(OZ)** of the drill string and axis **(OX)** perpendicular to **(OZ)**. Axis **(OY)** makes up the conventional orthogonal set of axes **[OX,OY,OZ]**. The sensor 10 is mounted at a

distance **r** from the **(OZ)** axis and the angle between the sensing axis **(OS)** and the rotational axis **(OZ)** is **m**.

In the second configuration, the sensor 10 is mounted in a plane which is parallel to the borehole axis **(OZ)** and with its sensing axis perpendicular to the axis **(OY)** and making angle **m** with the direction of the borehole axis **(OZ)**.

If the rate of rotation about the **(OZ)** axis is **w** and the components of **{V}** are **{VOZ}** along the **(OZ)** axis direction and **{VOXY}** in the **(OXY)** plane, then if the output from the sensor 10 for both configuration 1 and configuration 2 of Figure 1 is of the form

$$\mathbf{V(t) = VOZ.cos(m) + VOXY.sin(m).cos(w.t) + c}$$

where time **t = 0** when the axis **(OX)** is coincident with the direction of **{VOXY}** and **c** is constant for any fixed rotation rate **w**.

Thus, the sensor output at time **t** can be written:

$$\mathbf{V(t) = K1.cos(w.t) + K2} \qquad \ldots\ldots\ldots\ldots\ldots\ldots\text{(i)}$$

where **K1 = VOXY.sin(m)** and **K2 = VOZ.cos(m) + c** are constant if the vector amplitudes **VOZ** and **VOXY** are constant.

Sensor output integration

The integration of **V(t)** from any initial time **ti** to **ti + T/4**, where **T = 2.π/w**, the time for one revolution about (OZ), is

$$Q = \int_{ti}^{ti+T/4} K1.\cos(w.t).dt + \int_{ti}^{ti+T/4} K2.dt$$

Thus,

$$Q = \Big[(K1/w).\sin(w.t)\Big]_{ti}^{ti + T/4} + K2.T/4$$

or

Q = (K1/w).[sin(w.ti + w.T/4) - sin(w.ti)] + L

or

Q = (K1/w).[sin(w.ti + π/2) - sin(w.ti)] + L

or

Q = (K1/w).[cos(w.ti) - sin(w.ti)] + L .....(ii)

where **L** is a constant = **K2.T/4**.

Using equation **(ii)**, the integration of **V(t)** from an arbitrary time **t0** to time **t0+T/4** yields

Q1 = (K1/w).[cos(w.to) - sin(w.to)] + L .....(iii)

Using equation **(ii)**, the integration of **V(t)** from time **t0+T/4** to time **t0+T/2** yields

$$Q2 = (K1/w).[\cos(w.t0 + w.T/4) - \sin(w.t0 + w.T/4)] + L$$

or

$$Q2 = (K1/w).[\cos(w.t0 + \pi/2) - \sin(w.t0 + \pi/2)] + L$$

or

$$Q2 = (K1/w).[-\sin(w.t0) - \cos(w.t0)] + L \qquad \ldots(iv)$$

Using equation (ii), the integration of $V(t)$ from time $t0+T/2$ to $t0+3T/4$ yields

$$Q3 = (K1/w).[\cos(w.t0+w.T/2) - \sin(w.t0+w.T/2)] + L$$

or

$$Q3 = (K1/w).[\cos(w.t0+\pi) - \sin(w.t0+\pi)] + L$$

or

$$Q3 = (K1/w).[-\cos(w.t0) + \sin(w.t0)] + L \qquad \ldots.(v)$$

Using equation (ii), the integration of $V(t)$ from time $t0+3T/4$ to time $t0+T$ yields

$$Q4 = (K1/w).[\cos(w.t0+w.3T/4) - \sin(w.t0+w.3T/4)] + L$$

or

$$Q4 = (K1/w).[\cos(w.t0+3\pi/2) - \sin(w.t0+3\pi/2)] + L$$

or

$$Q4 = K1/w).[\sin(w.t0) + \cos(w.t0)\} + L \qquad \ldots.(vi)$$

Writing $K = K1/w$ and $\alpha = w.t0$, then equations (iii) through (vi) yield for the four successive integrations of $V(t)$

$$Q1 = -K.\sin\alpha + K.\cos\alpha + L \qquad \ldots\ldots.(vii)$$

$$Q2 = -K.\sin\alpha - K.\cos\alpha + L \qquad \ldots\ldots.(viii)$$

$$Q3 = K.\sin\alpha - K.\cos\alpha + L \qquad \ldots\ldots..(ix)$$

$$Q4 = K.\sin\alpha + K.\cos\alpha + L \quad \ldots\ldots\ldots(x)$$

Integration control

In order to control the sensor output integration, as just described, over four successive quarter periods of the drill string rotation, a train of **n** (with **n** any multiple of 4) equally spaced pulses per revolution must be generated. If one pulse P_0 of this pulse train is arbitrarily chosen at some time **t0**, the repeated pulses $P_{n/4}$, $P_{n/2}$ and $P_{3n/4}$ define times **t0+T/4**, **t0+T/2** and **t0+3T/4** respectively where the period of rotation $T = 2\pi/w$ and **w** is the angular velocity of rotation.

A suitable means for generating an appropriate control pulse train is described in US-A1-20020078745, which is hereby incorporated by reference.

In an alternative form of integration control, the sensor output waveform itself can be used with appropriate circuitry for defining the integration quadrant periods. In particular, the relatively low noise magnetic fluxgate output is well suited to act as input to a phase-locked-loop arrangement. Fig. 2 shows such an arrangement, successive output pulses defining the integration quadrants.

Rotation angle

Equations **(vii)** through **(x)** can be solved to yield angle α; there is a degree of redundancy in the possible solutions but, for example,

$$Q1 - Q2 = 2K.\cos\alpha$$

and

$$Q3 - Q2 = 2K.\sin\alpha$$

or

$$\sin\alpha/\cos\alpha = (Q3-Q2)/(Q1-Q2) \qquad ....(xi)$$

Since $\alpha = w.t0$, the angle $S(t0)$ between the axis **(OX)** and the direction of **{VOXY}** at time **t0** can be determined from equation **(xi)**, and the angle between **(OX)** and **{VOXY}** at any time **tm** measured from the arbitrary starting time **t0** is then

$$S(tm) = \alpha + w.tm = S(t0) + 2\pi.tm/T \qquad ....(xii)$$

<u>Magnitudes of vectors **{VOXY}** and **{VOZ}**</u>

Equations **(vii)** through **(x)** can be solved to yield the constant **L**:

$$L = (Q1 + Q2 + Q3 + Q4)/4 \qquad ......(xiii)$$

and the constant **K** can be determined from:

$$(K)^2 = [(Q1-L)^2 + (Q2-L)^2]/2$$
$$= [(Q3-L)^2 + (Q4-L)^2]/2 \qquad ...(xiv)$$

The magnitude of vector **{VOZ}** can be determined as

$$VOZ = (K2-c)/\cos(m) = (4.L/T - c)/\cos(m) \quad ....(xv)$$

provided that constant **c** is known.

The magnitude of vector **{VOXY}** can be determined as

$$VOXY = K1/\sin(m) = (K.w)/\sin(m) \quad ....(xvi)$$

<u>Inclination angle</u>

The inclination angle **(INC)** can be derived from the gravity vector **{G}** with the aid of a rotating accelerometer.

Referring to Fig. 3, where **(INC)** is the angle between the tool axis **(OZ)** and the gravity vector **{G}**,

$$GOZ = G.\cos(INC) \quad ....(xvii)$$

and

$$GOXY = -G\sin(INC) \quad .....(xviii)$$

The accelerometer output can be written as

$$VG(t) = GOZ.\cos(m) + GOXY.\sin(m).\cos(wt) + CP.\sin(m) + D.\sin(m) \quad ....(xix)$$

where **CP** is a centripetal acceleration term and **D** is a sensor datum term. The centripetal acceleration term **CP** is zero for configuration 2 and makes this the preferred configuration for mounting of the accelerometer.

Since **CP** is proportional to $\mathbf{w^2/r}$ and is constant for constant **w**, then clearly **VG(t)** is of the form

VG(t) = K1.cos(w.t) + K2(w)
(or **K1.cos(w.t) + K2** for configuration 2) ....**(xx)**

where **K1** and **K2(w)** are constants at constant angular velocity **w** in the case of configuration 1 and always constant in the case of configuration 2. the constants **K1** and **K2(w)** can be determined from the accelerometer output integrations as described above together with the angle **(Highside Angle HS = w.t)** between the axis **(OX)** and the direction of **{GOXY}**.

K1 = GOXY.sin(m) ....**(xxi)**
and
K2(w) = GOZ.cos(m) + D.sin(m) ....**(xxii)**
with
C(w) = CP.sin(m) + D.sin(m) ......**(xxiii)**
constant at constant angular velocity **w** (or for configuration 2 at all **w**).

A calibration procedure can be carried out to determine the values of **C(w)** for angular velocity values **w** (constant in the case of configuration 2) by calculating values of **K2(w)** with the rotation axis **(OZ)** horizontal when **C(w) = K2(w)**.

Thus, for any drilling situation with known angular velocity **w**, the vector components of the local gravity vector **{G}** can be determined as

$$GOXY = K1/\sin(m) \qquad \text{.......(xxiv)}$$

and

$$GOZ = (K2(w) - C(w))/\cos(m) \qquad \text{......(xxv)}$$

The inclination angle (INC) can then be determined from

$$\sin(INC)/\cos(INC) = -GOXY/GOZ \qquad \text{.....(xxvi)}$$

<u>Azimuth angle</u>

When using a rotating fluxgate, the azimuth angle (AZ) can be determined from a consideration of the magnetic vector {B}. What follows is applicable to both configuration 1 and configuration 2.

With reference to Fig. 4, it can be shown that

$$BOZ = BV.\cos(INC) + BN.\cos(AZ).\sin(INC) \qquad \text{....(xxvii)}$$

and

$$BOXY = (BN.\cos(AZ).\cos(INC)-BV.\sin(INC)).\cos(HS-MS) + BN.\sin(AZ).\sin(HS-MS) \qquad \text{.....(xxviii)}$$

or, with HS-MS = d a constant,

$$BOXY = (BN.\cos(AZ).\cos(INC)-BV.\sin(INC)).\cos(d) + BN.\sin(AZ).\sin(d) \qquad \text{.....(xxix)}$$

With **D** the fluxgate datum, the fluxgate output can be written

$$VB(t) = BOZ.\cos(m) + BOXY.\sin(m).\cos(w.t) + D.\sin(m) \qquad ......(xxx)$$

or

$$VB(t) = K1.\cos(w.t) + K2 \qquad ......(xxxi)$$

where

$$K1 = BOXY.\sin(m)$$

and

$$K2 = BOZ.\cos(m) + D.\sin(m)$$
$$= BOZ.\cos(m) + C \qquad ......(xxxii)$$

are constants which can be determined from the fluxgate output integrations as described above together with the angle **(Magnetic Steering Angle = MS = w.t)** between the axis **(OX)** and the direction of **{BOXY}**.

A calibration procedure can be carried out to determine the value of the constant **C** by calculating the value of **K2** while rotating about the direction of the axis **(OZ)** along which **BOZ = 0** when **K2 = C**.

Thus, for any drilling situation the vector components of the local magnetic field {B} can be determined as

$$BOXY = K1/\sin(m) \qquad ......(xxxiii)$$

and

$$BOZ = (K2-C)/\cos(m) \qquad ....(xxxiv)$$

With reference to Fig. 5, the horizontal component {BN} of the local magnetic field vector {B} can be represented by horizontal components {B1} and {B2} where

$$B1 = BOXY.\cos(d).\cos(INC) + BOZ.\sin(INC) \quad .....(xxxv)$$

and

$$B2 = BOXY.\sin(d) \quad ....(xxxvi)$$

The Azimuth Angle (AZ) can then be determined from

$$\sin(AZ)/\cos(AZ) = -B2/B1 \quad ....(xxxvii)$$

Also, the horizontal component of the local magnetic field can be determined from

$$BN = (B1^2 + B2^2)^{3/2} \quad ....(xxxviii)$$

and the vertical component of the local magnetic field can be determined from

$$BV = BOZ.\cos(INC) - BOXY.\cos(d).\sin(INC) \quad .....(xxxix)$$

Earth's rotation vector

Where it is not practicable to use a magnetic fluxgate, this may be replaced by a rate gyro as sensor.

With reference to Fig. 6, if the geographic latitude at the drilling location is **(LAT)** then the vertical component of the earth's Rotation Vector **{RE}** is

$$\mathbf{RV} = -\mathbf{RE}.\sin(\mathbf{LAT}) \qquad \ldots\ldots\ldots\ldots(\mathrm{xl})$$

and the horizontal component is

$$\mathbf{RN} = \mathbf{RE}.\cos(\mathbf{LAT}) \qquad \ldots\ldots\ldots(\mathrm{xli})$$

The magnitude of the cross-axis rate vector **{ROXY}** can be shown to be

$$\mathbf{ROXY} = (\mathbf{RN}.\cos(\mathbf{GAZ}).\cos(\mathbf{INC})-\mathbf{RV}.\sin(\mathbf{INC})).\cos(\mathbf{d}) + \mathbf{RN}.\sin(\mathbf{GAZ})\sin(\mathbf{d}) \qquad \ldots\ldots\ldots(\mathrm{xlii})$$

where **(GAZ)** is the gyro azimuth angle and **d = HS - GS** is constant.

Since **RN, RV, d** and **INC** are known and **ROXY** can be derived as discussed below, **(GAZ)** can be determined.

With the particular configuration where the rate gyro sensing axis is perpendicular to the drill string rotation axis **(OZ)**, the rate gyro output can be written

$$\mathbf{VG(t)} = \mathbf{ROXY}.\cos(\mathbf{w.t}) + \mathbf{D} \qquad \ldots\ldots(\mathrm{xliii})$$

where **D** is the rate gyro datum, or

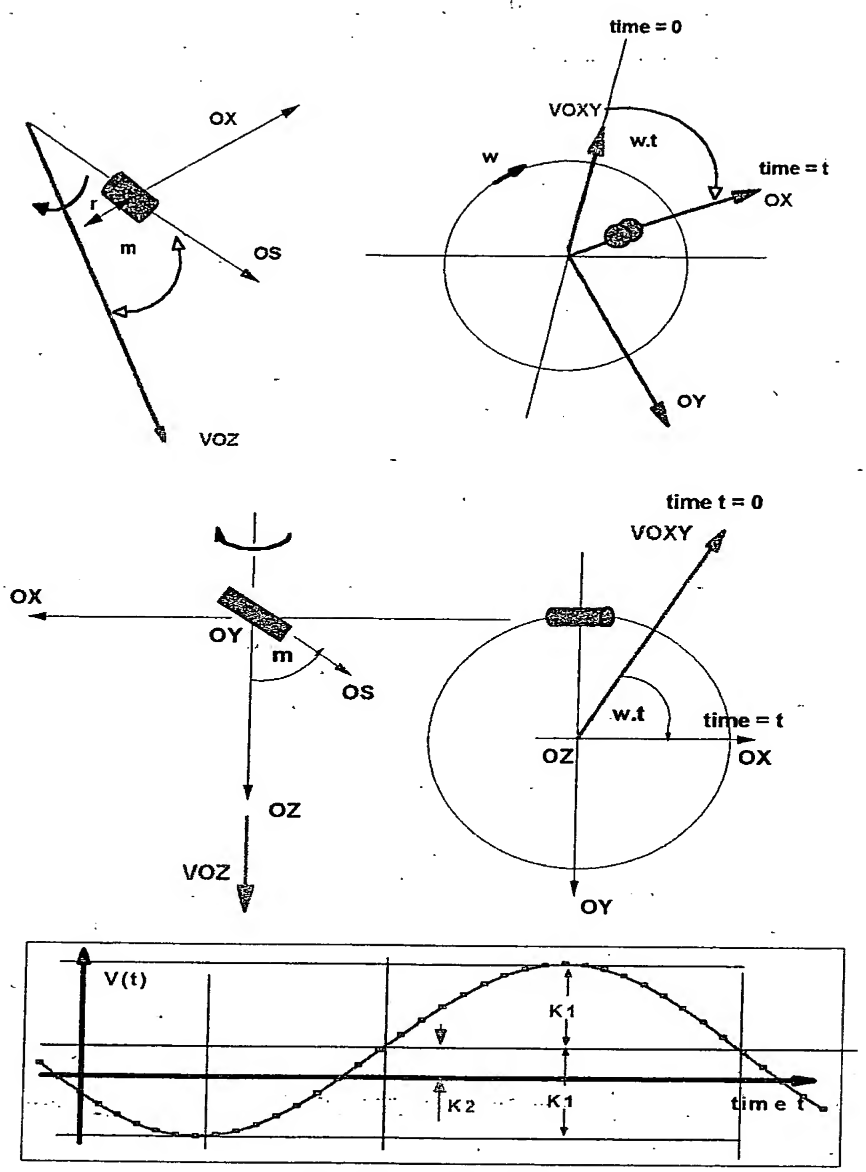
$$\mathbf{VG(t)} = \mathbf{K1}.\cos(\mathbf{w.t}) + \mathbf{K2} \qquad \ldots\ldots\ldots(\mathrm{xliv})$$

where the constant **K1** = **ROXY** can be determined from the rate gyro output integrations as described above together with the Gyro Steering Angle **GS** = **w.t** between **(OX)** and the direction of **{ROXY}**.

The variation in the Rate Gyro Datum makes it difficult to achieve satisfactory datum calibration in all circumstances. It is unlikely that Gyro Azimuth measurements should be attempted at high inclination angles. The use of the rate gyro is most likely with near-vertical boreholes in locations where magnetic azimuth measurements are unreliable (such as close to rigs) and the Gyro Azimuth **GAZ** is approximately equal to the angle **d**.

The present invention thus makes possible the measurement of a number of borehole-related parameters during rotation of a drillstring and using a reduced number of sensors. Modifications may be made to the foregoing embodiments within the scope of the present invention.

FIG. 1



Configuration 1

Configuration 2

$V(t) = K1.\cos(w.t) + K2$

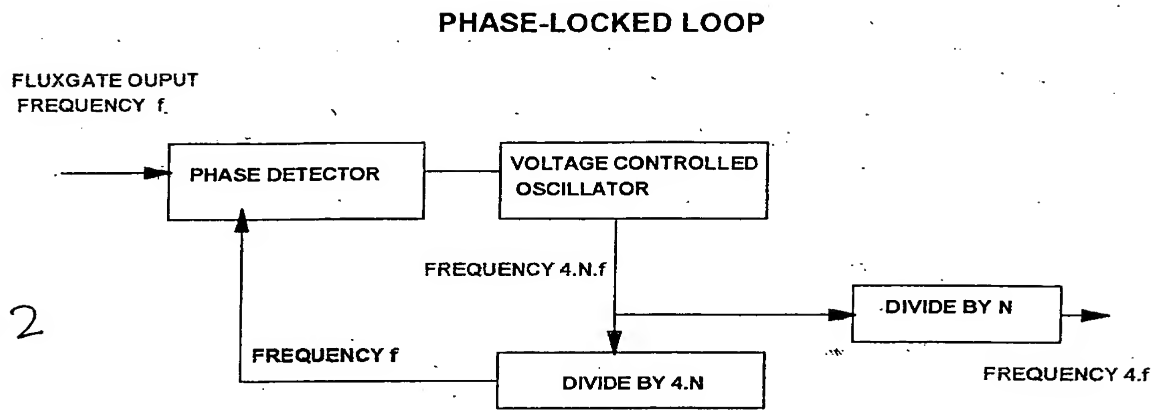
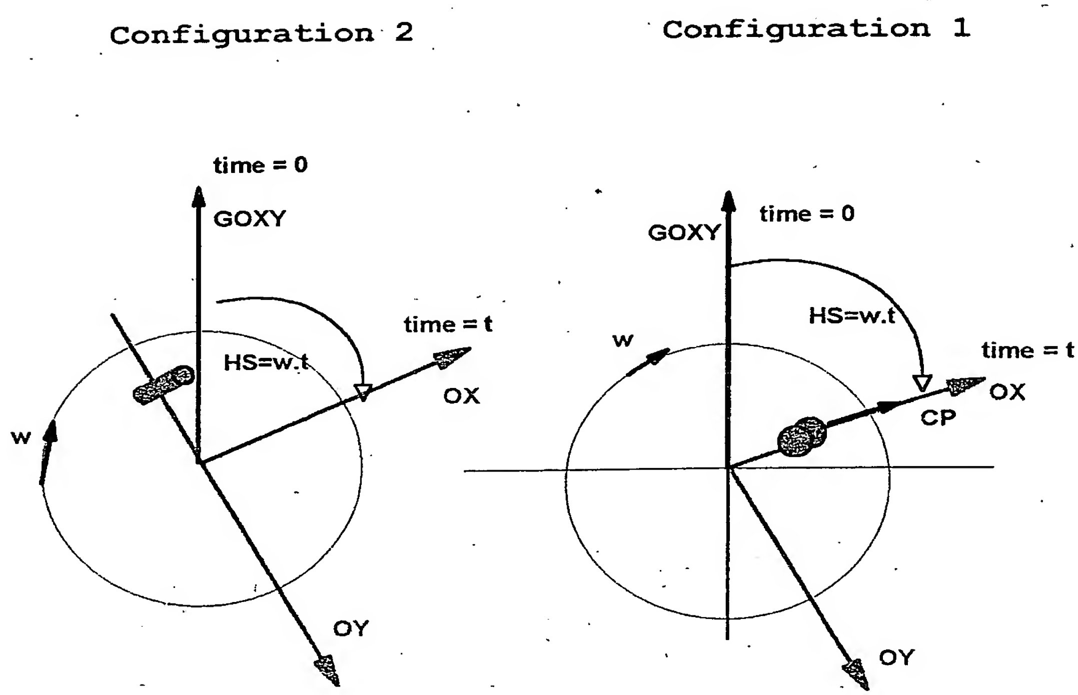

PHASE-LOCKED LOOP
FLUXGATE OUPUT
FREQUENCY f
PHASE DETECTOR
VOLTAGE CONTROLLED
OSCILLATOR
FREQUENCY 4.N.f
DIVIDE BY N
FREQUENCY f
DIVIDE BY 4.N
FREQUENCY 4.f

FIG. 2

FIG. 3


Configuration 2
Configuration 1
time = 0
GOXY
time = t
HS=w.t
w
OX
OY
time = 0
GOXY
HS=w.t
w
time = t
OX
CP
OY

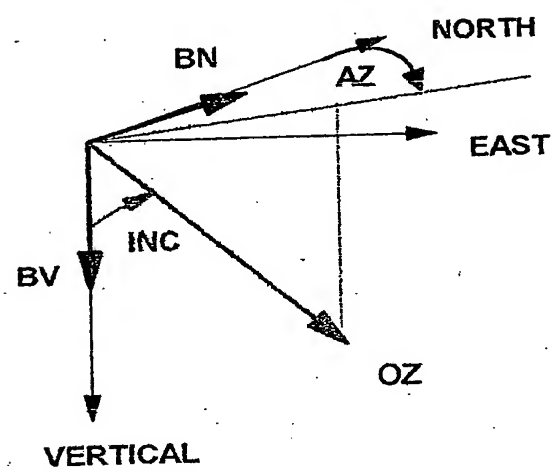


FIG. 4

FIG. 5

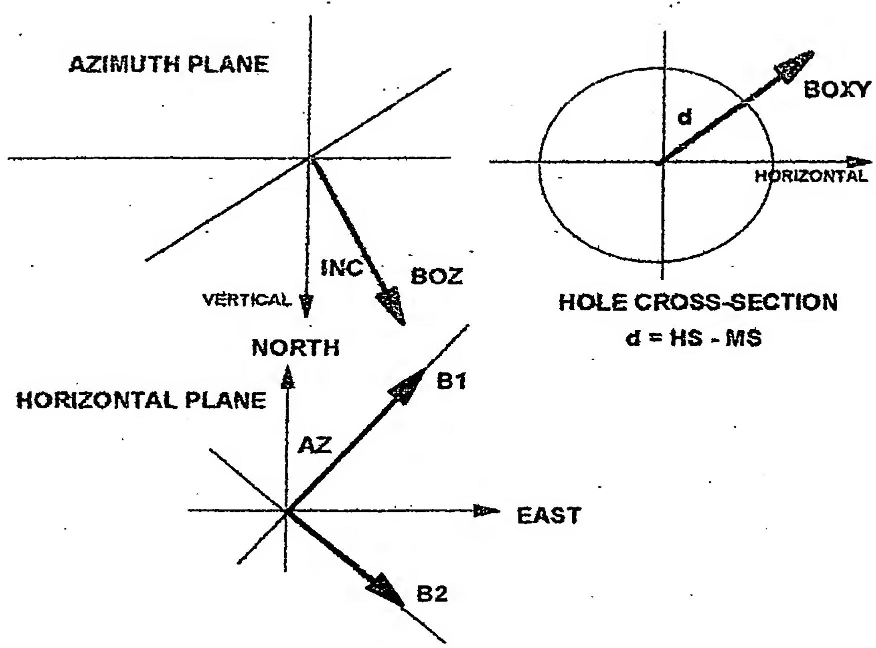
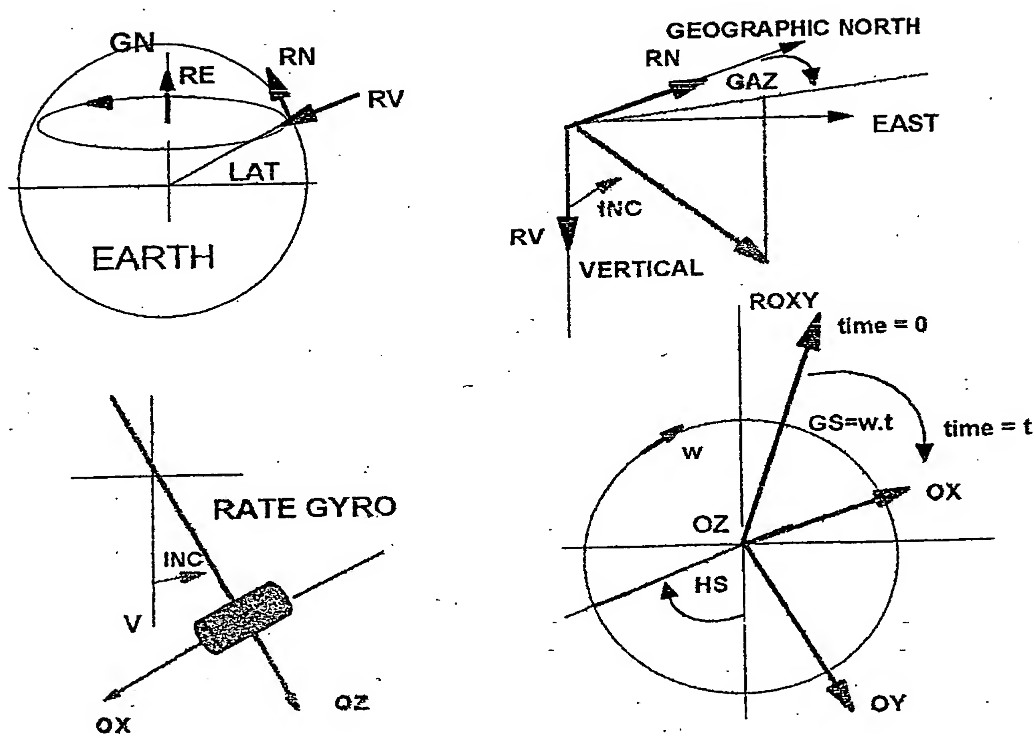

AZIMUTH PLANE
INC
BOZ
VERTICAL
BOXY
d
HORIZONTAL
HOLE CROSS-SECTION
d = HS - MS
NORTH
HORIZONTAL PLANE
AZ
B1
EAST
B2

FIG. 6


GN
RE
RN
RV
LAT
EARTH
GEOGRAPHIC NORTH
RN
GAZ
EAST
INC
RV
VERTICAL
ROXY
time = 0
GS=w.t
time = t
w
OX
OZ
HS
OY
RATE GYRO
INC
V
OX
OZ